

OPPOSING CONICAL PRELOADED ELASTOMERIC BEARING ASSEMBLY

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RELATED APPLICATION

This application incorporates by reference application titled "Installation of Internally preloaded Opposing Conical Elastomeric Bearing" invented by Neal Muylaert; attorney 15 docket BOEI-1-1006.

FIELD OF THE INVENTION

This invention relates generally to elastomeric bearings and specifically to opposed internally pre-loaded conical elastomeric bearings.

BACKGROUND OF THE INVENTION

20 A key component of a helicopter is the main rotor hub. It provides attachment of the main rotor blades during operation. Rotational power is delivered to the main rotor hub to provide rotational velocity to the blades in order to create aerodynamic lift. The main rotor hub must allow for rotational motion of the blades in the vertical (flap), horizontal (lead-lag), and axial (pitch) directions near the blade root attachment with the hub to accommodate 25 flight control authority and dynamic stability. Main rotor hub systems that accommodate these motions with discrete hinge mechanisms are referred to as fully articulated hubs. Through out the history of the helicopter, engineers have struggled to provide these rotational freedoms with bearing systems that can accommodate high frequency and high amplitude oscillatory motion under high trust loading created by the centrifugal force of the rotating 30 blades. Elastomeric bearings have become an industry standard for accommodating flap-

wise motion in articulated hub systems. These bearings are composed of elastomeric material that allows for shear compliance within the elastomer, and for rotational freedom while reacting radial centrifugal force in compression.

Elastomeric conical bearings are commonly used in bearing assemblies for helicopter 5 rotor systems to accommodate rotor motion. The bearing assemblies are axially preloaded to prevent the conical bearing elements from experiencing a resultant tensional load. Currently, mono-directional bearing elements are employed at each attachment site of the main rotor hub. FIGURE 1 depicts a view of a prior art articulated hub assembly 20a. The hub assembly 20a includes a tire bar 26 connected to a hub center body 22. The tie bar 26 is 10 connected to the center body 22 in a similar manner as disclosed in FIG.1, however, the bearing assembly 30a is substantially different. The bearing assembly 30 includes a pair of conical bearing elements 52 contacting the journal 28 on the bearing's inner surface 52 and the outer bearing surface is contained within an outer housing 42a. Each bearing element is a mono-directional single conical taper bearing having an elastomeric element 54 contained 15 within. The conical bearings are arranged such that the apex of the conical elements extends radially outward from one another. The bearing arrangement yields a force couple that extends from one bearing to the other. The force couple yields a bearing pre-load path 43 extending through the hub center body 22.

The prior art design creates an extended force couple resulting in a bearing pre-load 20 path extending through the main rotor hub center body. The hub center body must be designed to carry the extra loading. The extra design requirements add weight to the overall rotor hub reducing the aircraft's load capacity and fuel efficiency.

SUMMARY OF THE INVENTION

The invention provides a weight-reducing bearing assembly for rotary aircraft. An 25 opposed tapered conical elastomeric flap bearing assembly for rotary aircraft includes an outer housing having an outer surface and an inner surface. The outer surface is configured to mechanically connect the bearing assembly to the attachment sections of the hub center body. The inner surface is configured to receive a pair of opposed taper conical bearing elements. An inboard bearing element and an outboard bearing element are located within the outer 30 housing. The bearing elements are arranged in an opposed manner. An axial pre-load can be applied to the opposed bearing assembly wherein the resulting force couple bearing pre-load path is maintained entirely within the bearing assembly. Consequently, the weight of the main rotor hub is reduced increasing the efficiency of rotary flight.

The proposed invention provides a unique flap bearing arrangement by localizing the 35 pre-load within each flap bearing assembly and, thus, eliminates the necessity for the transfer

of the pre-load through the hub structure. The elimination of bearing pre-load through the hub structure can significantly reduce weight of the rotor hub assembly. Many components, including the bearing attachment flanges on the hub center body as well as the bearing housings can be configured to accommodate only the design flight and static loads without
5 having to carry the off axis bearing pre-loads. The unique design of the instant invention yields an approximate 6% - 10% weight reduction in the main rotor hub assembly.

BRIEF DESCRIPTION OF THE DRAWINGS

The preferred and alternative embodiments of the present invention are described in detail below with reference to the following drawings.

- 10 FIGURE 1 is a partially sectional isolated plan view of a prior art bearing assembly;
FIGURE 2 is an isometric view of an articulated hub assembly;
FIGURE 3 is a partially sectional isolated plan view of a bearing assembly of the instant invention;
FIGURE 4 is an exploded isometric view of the bearing assembly;
15 FIGURE 5 is an isometric view of the bearing assembly;
FIGURE 6 is an alternative isometric view of the bearing assembly;
FIGURE 7 is a partial sectional view of the bearing assembly; and
FIGURE 8 is a partial sectional view of the bearing assembly.

DETAILED DESCRIPTION OF THE INVENTION

20 FIGURE 2 depicts a fully articulated hub assembly 20 that includes a pre-loaded, opposed flap bearing assembly 30 that maintains a force couple bearing pre-load path entirely within the flat bearing assembly 30. The hub assembly 20 includes a plurality of rotor assemblies 24 radially attached to a hub center body 22. The articulated hub assembly 20 is designed to allow and to control the flap, pitch and lead-lag motion of an
25 aircraft rotor.

In a presently preferred embodiment, the rotor assembly 24 includes a tie bar 26. However, any other rotor attachment structure or assembly is considered within the scope of the invention. The tie bar 26 is a substantially cylindrical shaped element having a pair of radially opposed journals 28 at an end. Each journal is designed to receive the flap bearing assembly 30. The bearing assembly 30 extends over the journal 28 attaching itself to the journal 28. The tie bar 26 and bearing assembly 30 combination attach the rotor assembly 24 to the hub center body 22.

The flap bearing assembly 30 includes an inboard bearing element 32 and an outboard bearing element 34 contained within an outer housing 42 to form the bearing

assembly 30. The outer surface of the outer housing 42 is configured to attach the bearing assembly to another structure, for example, the main rotor hub 22. In a presently preferred embodiment, the outer housing 42 includes two pair of radially extending bearing flanges 36 configured to mate with a hub yolk 38 of the hub center body 22. However, any other structure or arrangement for attaching the bearing assembly to the rotor hub located on the outer housing 42 is considered within the scope of this invention, for example, a single pair of projections or molding the outer housing to the hub. A plurality of flange bores 60 align with yolk bores 40 allowing fasteners (not shown) to rigidly attach the structures.

FIGURE 3 depicts an isolated view of the hub assembly 20b of the instant invention. The tie bar 26 is attached to the hub assembly 20b via a pair of bearing assemblies 30b attached to the hub yolk 22 by attachment lug 58. The bearing assemblies 30b extend over and contact each respective journal 28. Each bearing assembly 30 includes a mated set of opposed, taper conical elastomeric bearing elements, 32 and 34, enclosed within an outer housing 42b. When preloaded in the axial direction, the opposed bearing assembly 30b limits the force couple to each individual bearing assembly 30b. According to the invention, the force couple is not passed through the hub center body. The force couple yields a bearing pre-load path 43b that remains entirely within each respective bearing assembly 30b.

FIGURE 4 depicts an exploded view of the flap bearing assembly 30b. The bearing assembly 30 includes an outboard bearing element 34 and an inboard bearing element 32 disposed within an outer housing 42b. The outer housing 42 includes a first section 45 and a second section 47.

The first section 45 includes a pair of radially extending flange sections 36. The flange sections 36 are configured to align with the hub yolk 38 (FIG. 1). An inner surface of the first section is shaped to receive the inboard bearing element 32. More specifically, an outer surface of the inboard bearing element 32 is bonded to the inner surface of the outer housing 42b in the first section 45. The bonding method is suitably any commonly known bonding method used in the art.

Disposed inside and adjacent the outer surface of the inboard bearing element 32 is an elastomeric element 54. The composition of the elastomeric element 54 can be any of the commonly employed elastomeric compositions, and is variable based upon the loading requirements of the employment environment. For example, an elastomeric element with a plurality of metal laminates is considered within the scope of this invention.

Positioned on an inner surface of the elastomeric element 54 is an inner race of the inboard bearing element 62. The inner race 62 includes a distal section 63 and a proximal section 65. The outer surface of the inner race 62 is tapered in the direction of the inner

bearing element 32. The inner surface of the proximal section 65 forms an axial bore 44 therethrough. The bore 44 is sized to receive the journal 28 through an open end 55 and extends into the distal section 63. The axial bore 44 terminates at an inner race closed end plate 53 located in the distal section 63. An outer surface of the distal section is substantially cylindrically shaped and configured to receive an inner race of the outboard bearing 64.

The outboard bearing element 34 is also a taper conical elastomeric bearing wherein the taper is in an opposing direction to the inboard bearing element 32. The outer surface 66 is configured to bond with the inner surface of the second section 47 of the outer housing 42b. The inner race has an open end to receive the outer distal section 63 of the inboard bearing inner race 62. At another end of the inner race of the outboard bearing 64 is an outer plate 56. Sandwiched between the outer race 66 and inner race 64 is another elastomeric element 54.

The outer plate and the inner race closed end plate 63 have a plurality of aligned bores extending therethrough. A tie bar attachment bore 46 is centrally disposed through each to receive a tie bar attachment lug (not shown). The tie bar attachment lug maintains the bearing assemblies 30 connection with the tie bar 26. Further, a plurality of coupler bores 48 area disposed through the respective surfaces. Each coupler bore receives a coupler lug 49 (FIG. 6) to maintain the spatial integrity between the inboard and outboard bearing elements. Further, a plurality of dowel bores extend through the respective plates, each bore receiving alignment dowels (not shown) extending from the journal end 29 (FIG. 1).

FIGURE 5 depicts an assembled view of the flap bearing assembly 30. The inboard bearing element 32 and the outboard bearing element 34 are coupled via a friction fit between the respective elements and the bearing coupler lugs 49. More specifically, the inner race of the outboard bearing 64 and the inner race of the inboard bearing 62 are frictionally mated upon insertion of the outboard bearing element 34. Additionally, the outer race of the outboard bearing 66 is bonded to the inner surface of the outer housing 42. Consequently, the outer housing 42 encompasses both the inboard bearing element and the outboard bearing in a single unitary assembly.

FIGURE 6 depicts the assembled flap bearing assembly 30. The bearing assembly includes an outer housing 42 surrounding the inner and outer bearing elements 32 and 34. The outboard bearing assembly 34 is pressure fit into the inboard bearing element 32 and then bonded between the outer race of the outboard baring 66 and an inner surface of the inboard bearing.

As discussed above, bearing coupler lugs 49 are disposed through the bearing coupler bores 48 connecting the outboard bearing element to the inboard bearing element.

Additionally, a tie bar attachment bore is axially located through the respective bearing elements and is in alignment with a respective bore in the journal end 29 (see FIGURE 1). An attachment lug 58 (not shown) is disposed through the tie bar attachment bore 46 and mechanically fastened to the journal end 29. Consequently, a bearing integrity redundancy is
5 created by to the two separate coupling structures.

The bearing assembly 30 is axially pre-loaded. In the preferred embodiment, a bearing assembly 30 axial pre-load of 8500-15,000 lb. range is desired. The pre-load helps to prevent the elastomeric elements of the bearing assembly 30 from tensional loading during operating conditions. However, any other pre-load is considered within the scope of this
10 invention. The axial pre-load can be applied through the coupler lugs 49, the tie bar attachment lug (not shown) or combinations thereof.

FIGURES 7 and 8 depict an isolated view of the opposed conical elastomeric bearings with and without pre-loading, FIG. 7, and with pre-loading, FIG. 8. A bearing gap 82 is located between the respective inboard and outboard bearing elements, 32 and 34
15 respectfully, prior to any axial pre-loading. As the axial pre-load is applied the bearing elements, 32 and 34, are brought together. The inner races, 62 and 64, frictionally engage one another and any space, or bearing gap 82, between the bearing elements, 32 and 34 is removed. The bearing elements, 32 and 34, combine within the bearing assembly 30 to carry the flap-wise motion of the rotor assembly 24.

20 While the preferred embodiment of the invention has been illustrated and described, as noted above, many changes can be made without departing from the spirit and scope of the invention. Accordingly, the scope of the invention is not limited by the disclosure of the preferred embodiment. Instead, the invention should be determined entirely by reference to the claims that follow.

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